

Examiner-Initiated Interview Summary	Application No.	Applicant(s)	
	10/650,166	GRATTON ET AL.	
	Examiner	Art Unit	
	GERALD L. SUNG	3741	

All Participants:

(1) GERALD L. SUNG.

(2) Justin Leach.

Date of Interview: 15 March 2009

Status of Application: After Non-Final Rejection

(3) _____.

(4) _____.

Time: 11:44 AM ET

Type of Interview:

- ☒ Telephonic
☐ Video Conference
☐ Personal (Copy given to: ☐ Applicant ☐ Applicant's representative)

Exhibit Shown or Demonstrated: ☐ Yes ☐ No

If Yes, provide a brief description: _____.

Part I.

Rejection(s) discussed:

N/A

Claims discussed:

1 and 16

Prior art documents discussed:

N/A

Part II.

SUBSTANCE OF INTERVIEW DESCRIBING THE GENERAL NATURE OF WHAT WAS DISCUSSED:

See Continuation Sheet

Part III.

- ☒ It is not necessary for applicant to provide a separate record of the substance of the interview, since the interview directly resulted in the allowance of the application. The examiner will provide a written summary of the substance of the interview in the Notice of Allowability.
☐ It is not necessary for applicant to provide a separate record of the substance of the interview, since the interview did not result in resolution of all issues. A brief summary by the examiner appears in Part II above.

(Applicant/Applicant's Representative Signature – if appropriate)

Continuation of Substance of Interview including description of the general nature of what was discussed: The amendments submitted 26 December 2008 pertaining to a rocket nozzle were discussed. In particular, the Examiner has brought into question the basis and support for the limitations "rocket nozzle" because the Examiner has found no explicit support in the Applicant's specification. The Applicant has responded with a request for Examiner's Amendment deleting the term "nozzle" and inserting "exhaust assembly." Antecedent support for the limitation "ablative rocket exhaust assembly" is provided, in part, by Applicants' originally submitted claims 1-15, which each recited "an ablative composite assembly." Antecedent support for the phrase "ablative rocket exhaust assembly," and specifically for the phrase "rocket exhaust," may also be found throughout the Application's Specification. See, for example, Applicants' Abstract stating that embodiments of the invention relate to ablative composite sub-assemblies capable of withstanding "the high temperatures (5000 degrees Fahrenheit) encountered in hot gas rocket exhausts"; Applicants' Paragraphs 0008-0009 generally explaining that the embodiments of Applications' invention represent an improvement over conventional exhaust valves used in applications such as for tactical missiles, [which] use inexpensive lightweight ablative composites for their construction" and which "have the disadvantage that they are prone to degrade at extremely high temperatures (Greater than 5000 degrees Fahrenheit)"; Applicant's Paragraph 0052 noting that "[t]he film adhesive 32 does not erode at the high temperatures (5000 degrees Fahrenheit encountered in hot gas rocket exhausts."